

RPS

Page ~~8~~¹¹, between lines 7 and 8, insert the following heading:

Remove { (N.E.)

--BRIEF DESCRIPTION OF THE DRAWINGS--

Page 15, replace the paragraph beginning at line 12, with the following rewritten paragraph/s:

--DETAILED DISCUSSION OF PREFERRED EMBODIMENTS

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Figure 1 shows skin panels 1, 2 friction stir butt welded together at 3 and having a stringer 4 bolted to the skin panels 1, 2 either side of the weld 3. A secondary load path is thus provided and the need for a butt strap removed.--

Page 15, replace the paragraph beginning at line 16, with the following rewritten paragraph:

--In Figures 2, 3 and 4 alternative structural assemblies for a wing skin or fuselage skin stiffened assembly are shown. Extruded panel stiffener members 5, 6 in Figure 2 are shown friction stir butt welded together at 3 with a butt strap 7 bolted in position to members 5 and 6 either side of the weld 3. Again by this means it will be seen that a secondary load path is provided.--

Page 16, replace the paragraph beginning at line 8 with the following rewritten paragraph:

--Figure 5A, B and C show alternative methods of attaching stiffeners to panel members. In Figure 5A extrusion 12 has a stiffening portion 13, friction stir butt welded to it at 3. In Figure 5B a friction stir butt weld 3 connects together two panel members 1,

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2 and also a T-shaped stiffener member 14. It will be observed that the weld 3 occupies the entire space between members 1, 2 and 14. In Figure 5C an alternative arrangement to that of Figure 5B is shown with a T-shaped stiffener 15 extending between panel members 1 and 2.--

Page 19, replace the paragraph beginning at line 6 with the following rewritten paragraph:

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--In Figure 12 part of a wing spar 35 is shown friction stir butt welded at 36 to an angled skin portion 37 which is in turn friction stir butt welded at 38 to a skin panel member 39. A separate rib post 40 is fastened in position to spar 35 and angled portion 37. This example teaches how a construction according to the invention works to the designers advantage in designing a complex structural joint assembly. In this example the portion of spar 35 shown comprises 7000 series aluminium alloy, the angled portion 37 comprises 2000 series aluminium alloy and the skin portion 39 comprises 2000 series aluminium alloy.--

IN THE CLAIMS

Please cancel claims 2-12, 15, 17, 19, 21-31 and 34-36 without prejudice.

REMARKS

In view of the cancellation of claims noted above, claims 1, 13, 14, 16, 18, 20, 32, 33 and 37-39 remain in this application.